

Drones and Drone Platforms



U.S. Navy Prototype Drone: RPB-Robotic Piloted Drone
Image Source: US Navy



Modern Navy X47B RPB Drone, aboard USS George H. W. Bush
Image Source: US Navy



MQ-25 Tanker Refueler Drone
Image Source: © Dezeen Limited, Boeing®

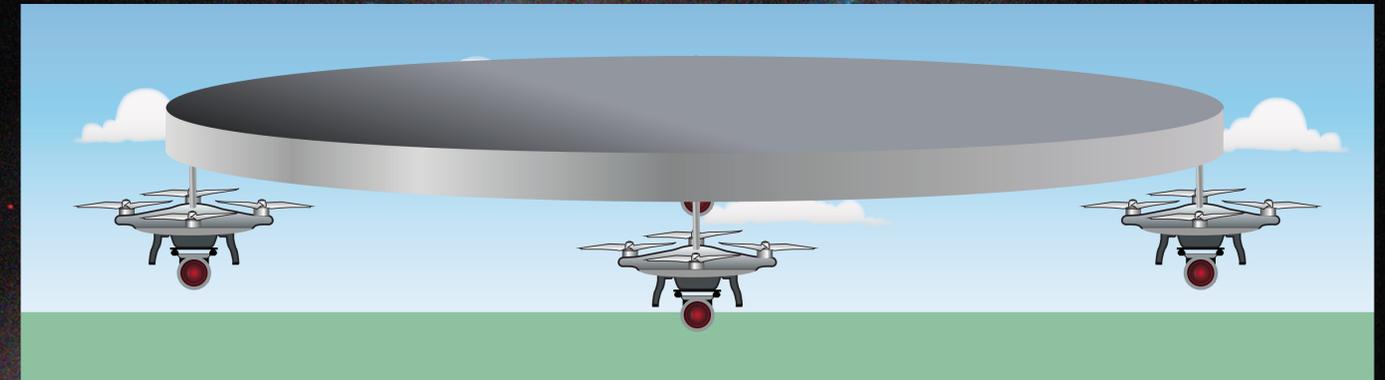
Speaker: John M. Dilorio (Main Room)
United Way of Greater Houston, 50 Waugh Drive, Houston, Texas 77007
Date: 2018
Time: 5:15 pm - 6:30 pm



Tutorial: "Drones and Drone Platforms"

The learning objectives for this tutorial are:

- The quality requirements for a Drone Platform.
- The commercial contractor customer requirements.



Design Criteria

- Jet engine, rotor engine, or particle thruster (thrust) of 30,000 lbs.
- Thrust/weight ratio of 1.2 (a computerized safety margin of 1.1)
- Advanced dirigible material and frame
- Limited Helium pockets
- Aerial flight stabilizers

Thrust/weight (raw) calculation: ~ (4 thrusters) times 30,000 lbs./x = 1.2 or weight = 100,000 lbs. or 50 tons.
Thrust/weight (Helium) calculation: ~ 2.5 factor or 100,000 lbs. times 2.5 = 250,000 lb. or 125 tons¹.
Thrust/weight (Jet) calculation: ~ (- thruster engine) times 2 (Piloted VTOL jets) = 60,000 lb. or 30 tons.
Thrust/weight (Safety) calculation: 50 tons plus 125 tons minus 30 tons/50 tons = 2.9 (RPV range)

¹ This takes in account the weight of the drone platform.



1 M Diam. Security Wi-Fi Drone

John M. Dilorio
has made presentations at
Space City Comic Convention,
ComicPalooza, in association
with Planet Forbidden Pictures
and NASA Science Faction.
Models — Prizes — Demonstrations

MQ9 Thrust Equation

$$T = \frac{\pi}{4} D^2 \rho v \Delta v$$

$T = \text{thrust [N]}$

$D = \text{propeller diameter [in]}$

$v = \text{velocity of air at propeller } \left[\frac{m}{s} \right]$

$\Delta v = \text{velocity of air accelerated by propeller } \left[\frac{m}{s} \right]$

$\rho = \text{density of air } \left[1.225 \frac{kg}{m^3} \right]$

